

## AMENDMENTS TO THE CLAIMS

Please amend the claims as follows. This listing of claims replaces all previously-filed listings.

1-10. (CANCELLED)

11. (CURRENTLY AMENDED-WITHDRAWN) A rotary-wing aircraft rotor system which rotates about an axis of rotation, comprising:

a rotor system having an N number of blades which rotates about an axis of rotation at a rotational speed of 1P, such that said main rotor system produces NP vibrations;

a sensor system which senses the NP vibrations;

a multiple of ~~independently rotatable~~ masses coaxially disposed with said rotor system;

a drive system ~~interconnected to each of said multiple of independently rotatable masses~~ to independently ~~rotate~~ spin each of said multiple of ~~independently rotatable~~ masses ~~relative to said rotor system~~ about said axis of rotation at an angular velocity; and

a control system in communication with said sensor system and said drive system, said control system operable to identify variations of the NP vibrations to control ~~an~~ the angular velocity of at least one of said multiple of ~~independently rotatable~~ masses to reduce the NP ~~in-plane~~ vibrations.

12. (WITHDRAWN) The system as recited in claim 11, wherein said rotor system includes a rotary wing aircraft main rotor system.

13. (CURRENTLY AMENDED-WITHDRAWN) The system as recited in claim 12, further comprising a generator driven by said main rotor system which powers an electric motor of said drive system, each of said multiple of ~~independently rotatable~~ masses driven by an electric motor, a phase of the voltage from said generator providing a phase reference to said control system indicative of a rotational speed of said main rotor system.

14. (WITHDRAWN) The system as recited in claim 11, wherein said drive system rotates at least one of said multiple of independently rotatable masses in a direction opposite to the rotational direction of said rotor system.

15. (WITHDRAWN) The system as recited in claim 11, wherein said drive system rotates at least one of said multiple of independently rotatable masses at an angular velocity greater than an angular velocity of said rotor system.

16. (WITHDRAWN) The system as recited in claim 11, wherein said control system utilizes a phase angle from a power source driven by said main rotor system as a phase angle reference to said control system.

17. (WITHDRAWN) The system as recited in claim 16, wherein said control system communicates with a sensor system interconnected to said main rotor system to provide feedback signals to said control system.

18. (CURRENTLY AMENDED-WITHDRAWN) A method of reducing vibrations in a rotary-wing aircraft main rotor system having N number of blades which rotate about an axis of rotation at a rotational speed of 1P such that the main rotor system produces NP vibrations comprising the steps of:

(A) independently ~~rotating~~ spinning each of a multiple of ~~independently rotatable~~ masses coaxially disposed about an axis of rotation of a main rotor system, each of the multiple of masses radially offset from the axis of rotation of the main rotor system; and

(B) controlling an angular velocity of at least one of ~~said~~ said the -multiple of ~~independently rotatable~~ masses relative to the main rotor system to reduce the NP vibrations generated by ~~of~~ the main rotor system.

19. (CURRENTLY AMENDED-WITHDRAWN) A method as recited in claim 18, wherein said step (A) further comprises:

(a) ~~rotating~~ spinning the multiple of ~~independently rotatable~~ masses ~~driven~~ at one revolution per cycle with the main rotor system.

20. (CURRENTLY AMENDED-WITHDRAWN) A method as recited in claim 18, wherein said step (A) further comprises:

(a) ~~rotating~~ spinning at least one of the multiple of ~~independently rotatable~~ masses in a rotation direction opposite the rotational direction of the main rotor system.

21. (CURRENTLY AMENDED-WITHDRAWN) A method as recited in claim 18, wherein said step (A) further comprises:

(a) ~~rotating~~ spinning at least one of the multiple of independently rotatable masses at three additional revolution per cycle in a rotation direction of the main rotor system.

22. (CURRENTLY AMENDED) A vibration isolation system for reducing vibrations in a rotating system rotatable about an axis of rotation, comprising:

a multiple of ~~independently rotatable~~ masses coaxially disposed about an axis of rotation of a rotating system, each of said multiple of ~~independently rotatable~~ masses ~~extends from a radial arm operable to spin about~~ radially offset from said axis of rotation;

a drive system to independently spin ~~said radial arms~~ each of said multiple of masses about said axis of rotation at an angular velocity; and

a control system in communication with said drive system to control ~~an~~ the angular velocity of ~~at least one~~ each of said multiple of ~~independently rotatable~~ masses to ~~reduce in-plane vibrations generated by~~ of the rotating system.

23. (WITHDRAWN) The system as recited in claim 22, wherein said rotor system includes a rotary wing aircraft main rotor system.

24. (CURRENTLY AMENDED-WITHDRAWN) The system as recited in claim 23, further comprising a generator driven by said main rotor system which powers an electric motor of said drive system, each of said multiple of ~~independently rotatable~~ masses driven by an electric motor, a phase of the voltage from said generator provides a phase reference to said control system indicative of a rotational speed of said main rotor system.

25. (CURRENTLY AMENDED) The system as recited in claim 22, wherein said drive system ~~rotates~~ spins at least one of said multiple of ~~independently rotatable~~ masses in a direction opposite to the direction of rotation of said rotating system.

26. (CURRENTLY AMENDED) The system as recited in claim 22, wherein said drive system ~~rotates~~ spins at least one of said multiple of ~~independently rotatable~~ masses at an angular velocity greater than an angular velocity of said rotating system.

27. (PREVIOUSLY PRESENTED) The system as recited in claim 22, wherein said control system utilizes a phase angle from a power source as a phase angle reference to said control system.

28. (PREVIOUSLY PRESENTED) The system as recited in claim 22, wherein said rotating system includes a rotary wing aircraft main rotor system.

29. (CURRENTLY AMENDED) The system as recited in claim 22, wherein each of said multiple of masses are mounted to an end of a said radial arm ~~defines a constant cross-section.~~

30. (CURRENTLY AMENDED) The system as recited in ~~claim 22~~ claim 29, wherein said drive system includes a multiple of electric motors, each of said multiple of electric motors spin one of said multiple of ~~independently rotatable~~ masses through said radial arm.

31. (CURRENTLY AMENDED) The system as recited in claim 30, wherein each of said multiple of electric motors are independently operated to independently spin each of said multiple of ~~independently rotatable~~ masses at an independent angular velocity.

32. (CURRENTLY AMENDED) The system as recited in claim 22, wherein each of said multiple of ~~independently rotatable masses spin about said axis of rotation to control said angular velocity~~ define an eccentric mass relative to said axis of rotation.

33. (CURRENTLY AMENDED) The system as recited in claim 22, wherein said multiple of ~~independently rotatable~~ masses are located only on one side of said rotating system.

34. (CURRENTLY AMENDED) The system as recited in claim 22, wherein said drive system spins each of said multiple of ~~independently rotatable~~ masses about said axis of rotation at an independent angular velocity.

35. (NEW) The system as recited in claim 32, wherein each of said multiple of masses are mounted to an end of a radial arm.

36. (NEW) The system as recited in claim 22, wherein said rotating system rotates at a rotational speed of 1P and generates NP vibrations